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AIR FORCE SYSTEMS COMMAND

WRIGHT PATTERSON AIR FORCE BASE OHIO



A USER'S MANUAL FOR THE
SEQUENCE ACCOUNTABLE FATIGUE ANALYSIS
COMPUTER PROGRAM

OCTOBER 1973

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FORWARD

This program was prepared by J. M. Potter of the Solid Mechanics

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Air Force Flight Dynamics Laboratory. This work was conducted in-house
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This Technical Memorandum has been reviewed and is approved.

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ABSTRACT

This report presents a detailed description of a computer program to calculate cumulative damage of notched structural members subjected to arbitrary spectra. The Sequence Accountable Fatigue Analysis computer program develops its sequence sensitivity by tracking residual stresses local to a notch throughout the spectrum of loads. Residual stress relaxation analysis is included to increase the generality of the results. An example spectrum and resulting cumulative damage analysis are illustrated.

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SYMBOLS

Tos	Residual stress
Jmax	Maximum local stress level
σ_{\min}	Minimum local stress level
σys	Yield stress
σ _{resEQ}	Equilibrium component of the residual stress
εt	Local strain, total
ε _e	Elastic component of total strain
ε p	Plastic component of total strain
S max	Maximum applied stress level
Smin	Minimum applied stress level
Smean	Mean applied stress level, (S _{max} +S _{min})/2
K _t	Elastic stress concentration Factor
D	Damage
I	Integer describing the level number
NED	Equilibrium period, number of cycles for the local stresses to return approximately to the equilibrium conditions following an overload
CI.	Residual stress relaxation constant
E1, E2	Relaxation function exponents
N	Number of cycles
E	Modulus of elasticity
N _£	Number of cycles of life at a given stress or strain cycling level
E #	Strain intercept at one reversal on a log ϵ_p -log life curve
· ·	Slope of the log $\varepsilon_{\mathbf{p}}$ -log life curve

SECTION I

INTRODUCTION

Cumulative damage analyses based upon the local stress-strain behavior at a notch appear to be reasonably successful in anticipating trends in fatigue life behavior of notched specimens subjected to spectrum loading (1-6). The type of behavior that usually occurs is that peak tensile loads tend to increase the fatigue life and peak compressive loads tend to decrease the life of notched structures compared to structures experiencing load spectra not having those peaks (5,6). Local behavior analyses, such as those developed by Smith (7) and Neuber (8), help to explain this phenomenon as being a result of the tensile peak load creating a compressive residual stress at the notch and, conversely, the compressive peak creating a tensile residual stress. The change in life occurs because the residual stress state modifies the subsequent damage accumulation rates.

The Sequence Accountable Fatigue Analysis computer program was developed to incorporate the local stress-strain approach with a recent residual stress relaxation analysis (6) in order to improve the sequence sensitivity of cumulative damage analysis. This technical memorandum presents the details of the resultant computer program and an example of its use. The correlation of predictions made with this analysis to actual results of tests experiencing spectrum loading is presented by Potter (9) and Potter and Noble (10).

SECTION II

PROGRAM OUTLINE

The Sequence Accountable Fatigue Analysis traces the stress-strain behavior local to a notch throughout an applied load spectrum and calculates the damage based on the local experience. The computer program is divided generally into the four parts or modules outlined in Fig. 1.

The basic input data for the material, specimen geometry, fatigue behavior qualities and spectrum, are developed in Module I. Module I is discussed further in Section III. Module II takes the input information and determines the local stress-strain behavior. Module III references the Range Pair Counting Method Subroutine to cycle count the local stress spectrum. Module IV determines the damage in the local stress-strain spectrum.

The basic analyses used in Modules II, III and IV are presented below.

Module II - Local Stress-Strain Behavior - The analysis used during the determination of the local stress behavior during the spectrum of loading is a combination of analyses developed by Smith (7), Newber (8) and Potter (6). Smith's simple analysis indicated that the residual stress could be approximated by assuming that the initial stress-strain behavior was elastic upon unloading following plastic flow. Thus, the residual stress could be calculated knowing the

maximum local stress and the maximum applied stress as in Eq. 1 and in Fig. 2.

$$\sigma_{res_{i}} = \sigma_{max_{i}} - K_{t}S_{max_{i}}$$
 (1)

The Sequence Accountable Fatigue Analysis computer program currently incorporates elastic-perfectly plastic stress-strain behavior. Therefore, σ_{\max_i} is equal to the yield stress. For the cycles immediately following the peak stress, the residual stress determined in Eq. 1 modifies the elastic solution as shown in Eqs. 2 and 3 (provided that the following maximum applied stress is less than S_{\max} and that there is no change in the residual stress due to a minimum applied stress causing reversed yielding).

$$\sigma_{\max_{i}} = \sigma_{\text{res}_{i-1}} + K_{t} S_{\max_{i}}$$
 (2)

$$\sigma_{\max_{i}} = \sigma_{\operatorname{res}_{i-1}} + K_{t} S_{\max_{i}}$$
 (3)

The analysis developed by Neuber (8) has been extended to cyclic loading by Wetzel (2) and Wetzel, Morrow and Topper (3) and used by many others primarily to determine local stress-strain behavior. It is used in this program only to calculate plastic strains occurring when the residual stress undergoes a step change. The plastic strain calculation routine is accessed only when the stress in Eqs. 2 and 3 exceed tensile or compressive yield stress levels, respectively. Figure 3 illustrates the calculation of the plastic strain.

The local stress-strain behavior, according to Wetzel (2) is related to the applied load by Eq. 4

$$\sigma \cdot \varepsilon = (K_t S_{max})^2 / E$$
 (4)

The plastic strain can be found by subtracting the elastic com-

$$\varepsilon_{\rm p} = \varepsilon_{\rm t} - \varepsilon_{\rm e} = (K_{\rm t} S_{\rm max})^2 / E \cdot \sigma_{\rm max} - \sigma_{\rm max} / E$$

Therefore, the plastic strain associated with S_{max_*} is given in Eq. 5.

$$\varepsilon_{p_i} = (K_t S_{max_i})^2 / E \sigma_{ys} - \sigma_{ys} / E$$
 (5)

If a residual stress existed prior to this plastic strain excursion, the plastic strain associated with that prior excursion is subtracted from Eq. 5 as shown in Eq. 6.

$$\varepsilon_{p_i} = (K_t S_{max_i})^2 / E \sigma_{ys} - (\sigma_{ys} - \sigma_{res_{i-1}})^2 / E \sigma_{ys}$$
 (6)

A similar calculation is made for plastic strains occurring during the minimum stress peak.

In the analysis developed by Potter (6) the residual stress cyclically relaxes toward zero or an equilibrium residual stress as shown in Fig. 4 according to Eq. 7.

$$\sigma_{\text{res}_{N=1},2,...} = (\sigma_{\text{res}_{N=1}} - \sigma_{\text{res}_{EQ}}) \exp(N/N_{EP_1} \ln(0.1))$$
 (7)

The N term, the Equilibrium Period, is dependent upon the applied stress and the Residual Stress Relaxation Constant.

$$N_{EP_i} = (C1/(K_t S_{max_i}^{E1} \cdot K_t S_{mean_i}^{E2}))$$
 (8)

The Residual Stress Relaxation Constant, Cl, has not yet been experimentally defined but should be a constant for a material.

Module III - Cycle Counting Method

After the local stress and plastic strain behavior is calculated,
the stress spectrum is Range Pair Counted using a computer program
developed by Tischler. (11)

Module IV - Damage Calculation

Damage is calculated separately for the plastic strain excursions and the elastic stress spectrum. The damage is determined from the conventional $D = \sum_{i=1}^{n} C_{i}$ calculation. Damage from each of the plastic strain cycles is determined from the Coffin-Manson (12) form

$$D_{i} = 1./N_{f_{i}} = 1./(\epsilon_{p_{i}}/\epsilon_{f_{i}})^{1/c}$$

Damage from the elastic stress cycles is determined in a similar manner. The maximum and minimum local stress levels are sequentially compared to unnotched S-N data in a Modified Goodman Diagram format. Damage is summed, and failure of the coupon is defined as the event occurring when the summed damage equals unity.

SECTION III

INPUT DATA REQUIREMENTS

In general, each spectrum analyzed will require slightly different programming in order to get the load history into a usable format for the core program. The basic program requires a certain family of information before any analytical predictions can be made. Appendix I contains a program listing for the Sequence Accountable Fatigue Analysis. The subroutine CORE which accesses the subroutines having to do with RPCM, the Range Pair Counting Method, contains the basic analysis. Subroutine SAL reads the data input and then references subroutine CORE. The subroutine SAL shown is one in which a block of cycles is repeated with optional cycles. A list of the input data cards and the resulting analysis is given in Appendix II.

The specific data requirements are given below.

- 1. Stress-Strain Behavior The stress-strain behavior is presumed to be elastic perfectly plastic with the tensile yield stress being equal to the compressive yield stress. The yield stress value used is an average of the monotonic behavior generally being above the 0.2% yield value and below the engineering ultimate strength.
- 2. Residual Stress Relaxation The residual stress relaxation
 behavior of Eq. 7 and 8 is characterized by Cl, the Residual Stress
 Balaxation Constant and El and E2, the relaxation equation exponents.
 The Residual Stress Relaxation Constant, Cl, has not yet been adequately
 determined. It should be a material property if the relaxation function

is correct and must be assumed. A reasonably accurate estimate of the Residual Stress Relaxation Constant for aluminum material falls in the range of 5-20 x 10⁶ (cycles) (Ksi)². Further experimentation on the part of the analyst should develop a Cl usable for his set of conditions until actual measurement of residual stress relaxation behavior defines the relaxation function and constants. At present El and E2 are considered to be equal to 1.0.

- 3. Specimen Geometry The elastic K_t value (if available) is entered into the analysis. If that value is not available then an estimate from some other method may be used. In certain cases, a value may be determined from a constant amplitude fatigue test of a similar structure by fitting several values of K_t to the analysis and determining the best correlation as is done with the K_f solution. Once a stress concentration factor, K_t, is determined for a specimen, that value is not changed from test-to-test of the same coupon configuration,
- 4. Load Multiplier Different spectra are presented for analysis in different manners. Some data are presented in percent of maximum stress, others in terms of nominal stress, and others in terms of hending moment. The value of the load multiplier defines the nominal stress history.
- 5. Cumulative Damage Analysis The damage from the range-paired elastic stress spectrum is determined by calculating a simple $\frac{n}{N}$ value for each level and accumulating the total. The N_{f_i} value is determined from unnotched coupon S-N data in the Modified Goodman Diagram format.

The program requires the input of four second order equations describing the maximum and minimum stress levels at lives of 10⁴, 10⁵, 10⁶ and 10⁷ cycles. The coefficients of the equations are derived by least square fitting the S-N data presented in the form of Eq. 9.

$$S_{\text{max}} = A(I)S_{\text{min}}^2 + B(I)S_{\text{min}} + C(I)$$
 (9)

The A, B, and C coefficients for several typical materials are presented in Appendix IV. The S-N data shown was derived from various sources but usually from the MIL-HDBK-5A (13). The C coefficients correspond to the maximum stress level at zero to maximum applied stress conditions on the unnotched coupons.

The damage from the plastic strain cycles is determined using the Coffin-Manson relation to calculate the N $_{\rm fi}$ value. The conventional plastic strain intercept at one reversal and the $\epsilon_{\rm p}$ - life slope values are used in the analysis. Specific measured values from the literature are used when available and typical values when they are not available.

6. Analysis or Test Spectrum - The last information needed is the order and magnitude of application of the spectrum used in the test.

SECTION IV

OUTPUT OPTIONS

The Computer Program prints the following output in the process of the analysis.

- 1. Maximum and minimum applied stress and local stress response through the spectrum. Also printed out is the residual stress, equilibrium stress, applied cycles, and the equilibrium period.
- The elastic local stress history as input into the Range Pair Subroutine and the resulting Range Paired spectrum.
- 3. The plastic strain occurrence during the spectrum and the damage associated with each strain reversal.
 - 4. The accumulated damage associated with the plastic strains.
- 5. The Range Paired elastic stress spectrum and the damage associated with each level.
- 6. The accumulated damage associated with the current block of loading including the plastic strain damage and the total damage since the initiation of cycling.

At the option of the analyst, he can print out all the above items or only two. The IPRINT value controls what data is printed.

If IPRINT = 1, all six items are printed for each flight or block.

If IPRINT = 2, all items except 2. above are printed. /

If IPRINT = 3, only items 4. and 6. above are printed.

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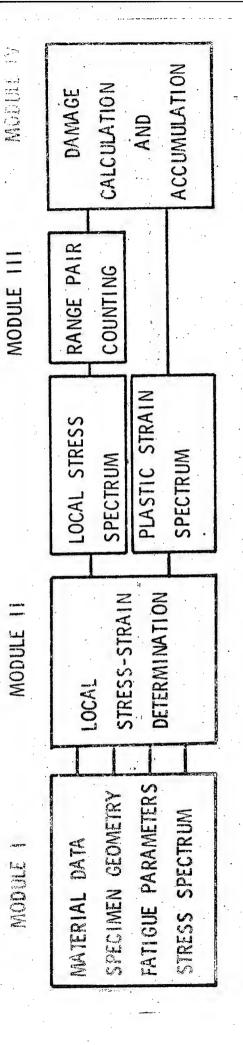
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PROCEDURE USED IN THE SEQUENCE ACCOUNTABLE FATIGUE ANALYSIS FIGURE 1.

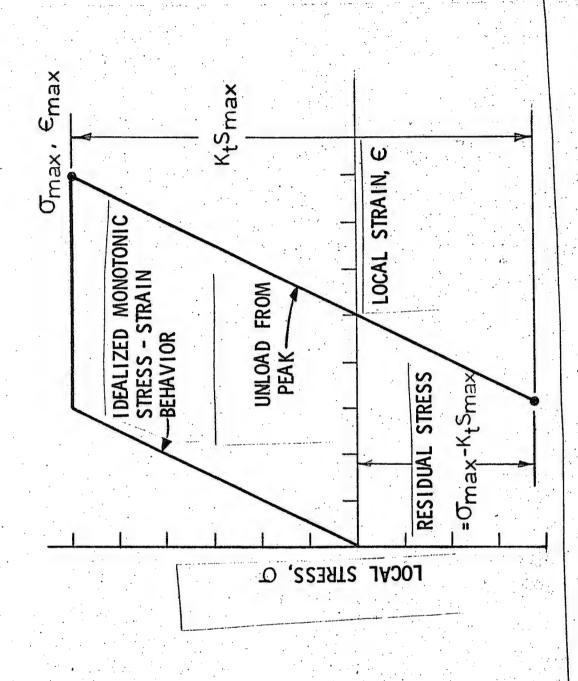


FIGURE 2. METHOD OF DETERMINING THE RESIDUAL STRESS FOLLOWING A PEAK LOAD

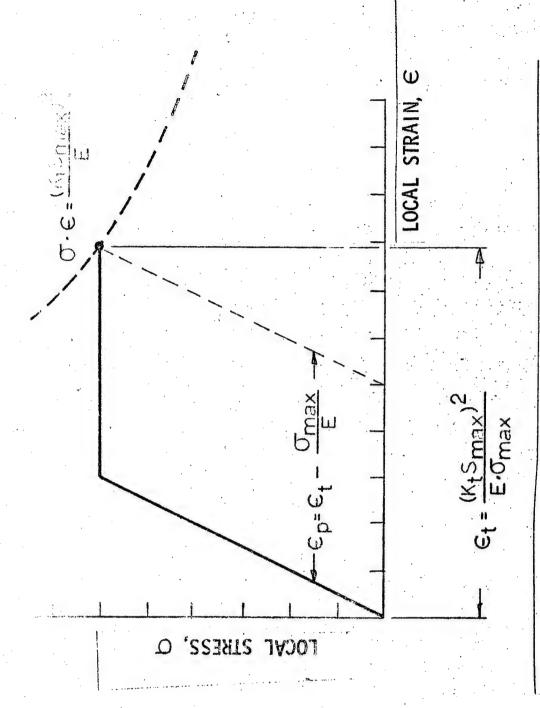
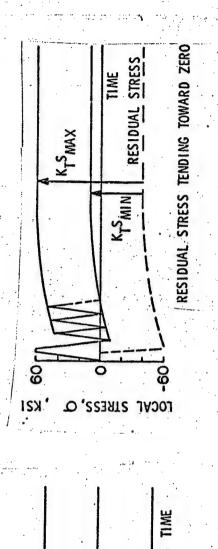


FIGURE 3. METHOD OF DETERMINING PLASTIC STRAIN LEVELS



K_TSMAX

-OVERLOAD

KTSMIN

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FIGURE 4. LOCAL STRESS RESPONSE FOR APPLIED CONSTANT AMPLITUDE LOADING WITH RESIDUAL STRESS RELAXATION

LOCAL STRESS RESPONSE

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SUBROUTINE	CORE	1 TO 4 25 0 TO WITH \$100 000			
		TELLO MICCONO NIL COO	03/15/13 B	U9.20.05. PAGE	2
	210 J3=-1 60A=ASMAY		CORE	57	
	BAB=TYS-RES(I-1)		CORE	78	
2	1F (838.3E.TYS) 60 TO 214 .	IF (BBB, SE, IYS) 60 YO 214.	CORE	60	
		MOD*TYS1-898*89877ELMOD*TYS1	CORE	62	
			CORE	63	
37			CORE	65	
			CORE	66	
			CORE	89	
2 02			CORE	70	
	290 SIGNAX(IRAIN) = PES (I) + ASMAX SIGNIN (IRAIN) = PES (I) + ASMIN	×	CORE	72	
Andreas and the state of the st	RNCYC(IRAIN) = ENN(J) IF (ASMAX, LE, TYS) GO TO 110		CORF	73	** The second se
7 54	EORES = 4 SMAX+TYS		CORE	75	
110	IF (ASM	30	CORE	77	
02.71	: :		CORE	79	and make the statement of the statement in the statement of the statement
-	0 DIF=RES(I)=CORES		CORE	81	
2	CALCULATE RELAXATION FUNCTION	NO LLON	CORE	83	
5	AND		CORE	85	the party of the same of the s
	ABMIN=ABS(ASMAX)		CORE	86	
	ABMEAN=ABS (ASMEAN) TF(ABMAX-1.		CORE	88	
06	IF(ASMEAN,LT.1.) ABMEAN=0.	5-2	CORF	06	
\$ 1	ABM=49MIN GO TO 446		CORE	31 92	
944	4 ABMEASMAX 6 ENEP=51/(ABM**F1*ABMFAN**E3		CORE	46	
95	TF (IPRINT GE.		CORE	95	
	* RES(1) - EORES FRM(3) STMIN	SIMAX (J), STMIN(J), SIGMAX (IRAIN), SIGMIN(IRAIN),	CORE		
350		(,=15.8,5X,16)	CORE	86 86	
100			CORE	100	
	GALCULATE RESIDUAL STRES	SS RELAKATION		102	
	I RAIN=IRAIN+1		ă. P	104	
105	60 19 350		CORE	105	
350	60 10 550 F (ABNIF 11 5-1) CO IN EED			107	
370	IF (1000, *ENN(J), LT.			108	the contraction of the contract of the contrac
110 330	NFLAGED	no.	-	11	and a construction can be seen to the company of th

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06.	TF (DUMNY -= NEP) 470,460,460	CORE 1	55
115 450	nummy=0JMMY/2.	CORE	7.7
	NFL43=NFL45+1 GO TO 450	CORE 1	18
.0.2%	CYCINT=DUMMY/10.		19
120	DECK-FLOAT (K.)	CORE 1	21 22
	F (K, =0.15LM) - DEON F (K, =0.15) F (K, =0.1		24
Det.			25 S
125 +90	2.393*5		27
015	IF(NFLAS-E0-0) 69 TO 530 NFLASS-MFLAG-10		259
	00 520 <=11,4F1462	CORE	30
130	EN(K) = 2, *0UMMY EX(K) = 5x0(-2,303*EN(K-1)/ENED) + EXP(-2,303*EN(K)/ENEP)		10.1
and the control of th	*DU4MY	CORE	34
320			25
135	A VSGMX (4) = 4SM4X+EDPES+01F+EX(F)/Z,		37
		COO	38
	SIGNAY (IRATH) = AVSGMY (K)	CORF	40
140	IF (K, = 0, 1) 50 TO 543	COPE 1	41
05:	OT SATMITTS	CORE	747 (43
250	(F7.2,1X),16X,F5.2,17X,F15.8)	CORE	99
145	CONTINJE	CORE	
956	CONTINUE	CORF	67
09:	CONTINUE RES(I)=EQRES+DIF*EXP(-2.303*ENN(J)/ENEP)	CORE	051
150 570		CORF	151
	LINITALIA	CORE	200
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155		STO	2
0 0	CYCLE COUNTING TECHNIQUE		158
c ¢	经通过的 计多数 计多数 计多数 计多数 计多数 计多数 计多数 医多种		091
160	CALL SUBROUTINE TO RANGE PAIR COUNT SPECTRUM	· 1	161 162
9	104		153
	RPSM(IN)		165
100	651 64 61		

591	CONTINUE		167	
592	KPHAX=IN CONTINUE	CORE	169	
170	***************************************	CORE	170	
0 0	MODULE IV	CORE STO	172	
0.0	DAMAGE ACCUMULATION CALCULATION	CORE	174 175	***
175	化热液促热 种致物 化双次试验 经发出 大大 医牙内皮 计大声传统 计电大道电话 计对比电话 医牙术 医大性 医阴道 医牙囊性 医生物性 医生物性 计计算机 计分析	CORE	176	
	T0 552	CORE	178	
180	FORMATIVIOSAL STRESSES AND PLASTIC STRAINS WIRESULTING FATIGU *E LIFE//10x,44STEP,10x,144PLASTIC STRAIN,10x,104MAX OR MIN,15x,6HD	CORE	180	
552	es quint, desprise mani, e com man mana en parte e dante desprise de commente com mana e man esta com		182	
	CALGULATE DAHAGE FROM PLASTIC STRAIN CYCLES	CORE	184	
185	SUMDEL≠0.	CORE	186	
And the state of t	. 00 531 JKL=1, WLEVEL	CORE	188	
190 532	IF(PLSTRA(JKI) 532,531,533	CORE	190	
233	PESTRATJKL)=AA*PESTRA (JKL) CYCLES=(PLSTRA (JKL) /EPSD) **GOFMAN	CORE	192	
77	SUMNO=SUMNC+DAM	CORE	194	
195	DEC.	CORE	196	
535	,0 A *	CORE	198	
200		CORE	200	
219	WRITE'16, 2197UKL, PLSTPA(UKL), DAM FORMAT (10x,14,12x, F10,5,15x,3H4Ax,15x,E14,6)	CORE	202	
231	GONTINUE WRITE(6,541) SUMDEL	CORE	204	
205 541	9X,29	CORE	206	
13	WRITE(6,13) FOPMAT(/16x,15H SIGMAX SIGMIN,18X,6H RNCYC,20X,25H CYCLES	CORE	208 209	
210 536	CONTINUE.	CORE	210 211	
9	CALCULATE ELASTIC CYCLE DAMAGE FROM LEAST SOUARE FITTED S-N DATA	CORE	212	
0	(MO)IFIED GOODHAN DIAGRAY FORMAT)	CORE	214 215	
215	00 600 JKL=1.KPMAX TTYS=TYS/5.	CORE	216 217	
	X=SISHINIJKL) Y=SIGMAX(JKL)	CORE	218 219	
220	IF(Y-X.LT.1.6*TYS) GO TO 310 CYCLES=10.**4.	CORE		
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3 C 1 CC TTVC1 CO 1 1 3 3 2 5		50 TO 340		00 330 4=1+4 0 4N - A 4N * * * * 2 ± 0 (N) # X ± C (N)		TOTAL TAILS	F IR (7) *Q(4) 339, 338, 33		3 954 = 485 (2 (41)	IFIA937.15.4834 1 63 TO	EXPO = 4, + 8 (4) / (P (4) - 8 (5))	60 10 339	EXF0=7,+R(7)/(R(6)-2(7))	6C TO 339	SUMMER (+) +0(5) +4(6) +9 (7)	101 ではおしいでしてもいましても一次川がマスロの	おものとも(の)光子のとく(す)光明の気の(の)の一名の大きの(の)の一名の大きの(の)の一名の大きの(の)の一名の大きの(の)の一名の大きの(の)の一名の大きの(の)の一名の「から)の「から)の「から)	CHARLES 40 CALL AD CALL AD CALLED	CA STORY DECK STRUCKS	DEL1=4, * SU4R2* SUMR24-4, * SU	0512=SUMR*SUM92*SUM33-SUM	DEL3=50:12*SIJN22*SUN23-S	001=20.**SUNKO**SUNK4-20.**SUNG4*	DOWNERS TO STAND THE STANDAY OF THE	EXPO = (001+002+003) /(0 = (1+0 = (2+0 = (3)	CYCLES=10.**EXPO	IF (EXPO.LE.4.) CYCLES=10.	COMMUNICATION OF COMMUN	SUMBEL = SUMBEL + ENACYC (JKL)		WeIT= (6, 539) Y, X, RNOYC	CONTINUE	WRITE(6, 593) SUMDEL	69X,21H DANAGE	FORMATCAGOY ASH TOTAL	מאל דמו ימי				
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Continuity Con	-		GIVEN LOAD SPECTS			269 270		
1			MAGGOOG		CORE	271		
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15)	SISMAX	PEAKS OF THE INOUT LOAD	NPKS + KK	i	276		
15		×	WHISH THE PROGRAM WIN	KCLUULNG INPUT		278		
15			OF THE	NPKS +	CORE	273		
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Total Common C			TRUM	2*NPKS	CORE	282	. 9	
20 0 0 0 0 0 0 0 0 0			OF FLEMENTS IN	NPKS +	CORE	283		
25 0 0000 0000 0000 0000 0000 0000 0000			THE CYCLES OF	+ SMAN .	CORE	285		
25			OF FLEMENTS OF	+ SHON	CORE	287		
STAVE VALLES OF NSTEPLJJ SUCH GORE		٠	YSIS SPECTRUM		CORE	288	A COMP CAMPING CAMPING CONTRACTOR OF A CAMPING	
1	:		AND VALUES OF NSTEP	_	CORE	2 2 2 2 3 3		. *
COMMON/MOEAL/RMCYC(7201) KPMAY, IPDPINI			SIGMAX(J-1) = SIGMIN	AND	CORE	291		
30 COMMON/ASAL/RNCYC(200), KPHAK, IPPINI COMMON/ASAL/RNCYC(200), LT. (1941) (2041) (2001) COMMON/ASCCASIC (190), LT. (1941) (190) COMMON/ASCCASIC (190), LT. (1941) (190) COMMON/ASCOTCLE (190), LT. (190), LT. (190) COMMON/ASCOTCLE (190), LT. (190), LT. (190) COMMON/ASCOTCLE (CORE	293		
COMMONYMOEOEZASTEP(3D3);LR;KMIY;K31 CORE	30	COMMONIAGE			CORE	295		
35 COMMONYGOGOCCE (900.2), RNECYC (900), NNSTEP(900) 36 COMMONYGOGOCL.LIND CORMONYGOGOCL.LIND DO 8000 I = 1,NPKS 8000 NSTEP (I) = I NATERIA: GE-25 GO TO 103 RAITE(5.20) NPKS 20 FORMAT (IHO.60HTHE MUM9ER OF PEAKS OR VALLEYS IN THE IMPOT LOAD SPE CORE 15 FORMAT (IHO.60HTHE MUM9ER OF PEAKS OR VALLEYS IN THE IMPOT CORE 15 FORMAT (IHO.60HTHE MUM9ER OF PEAKS OR VALLEYS IN THE IMPOT CORE 16 FORMAT (IHO.60HTHE MUM9ER K) 17 FORMAT (IHO.60HTHE LOAD SPECTRUM - PULL OUT THOSE PEAKS AND VALLEYS CORE 10 FORMINJE RISCESS THAN 1:0 CORE CONTINJE COUNTER K IS LESS THAN 1:0 CORE		COMMONZADEC		TANDS. TANDA.KIND	CORE	297		
35		SYSK NOMMOS	ZCYCLE (900,2) RNECYC (900) NNSTEP		CORE	299	***************************************	
9999 NPUNCH = 0 008E 8000 05 8000 1 = 1,NPKS 6000	35	I NCISNEWIO	SAVE (99), TITLE (8)		CORE	301		
DO 80 00 0 1 = 1,NPKS 0000 NSTEP(I) = 1,NPKS LF (PRINT.GE.2)50 TO 103 RRITE(6,20) NPKS RRITE(6,20) NPKS LOTEUM = 15/7) RRITE(6,20) NPKS LORE LORE LORE LORE CORE CORTINJS CONTINJS CONTINJS CONTERV INCLESS THAN 1:0 CORE COUNTER K IS LESS THAN 1:0 CORE CORE CORE CORE CORE CORE CORNIER K IS LESS THAN 1:0 CORE CORE CORE CORE CORE CORE CORNIER K IS LESS THAN 1:0 CORE CORE CORE CORE CORE CORE CORNIER K IS LESS THAN 1:0 CORE CORE CORE CORE CORE CORE CORNIER K IS LESS THAN 1:0 CORE CORE CORE CORE CORE CORE CORNIER K IS LESS THAN 1:0 CORE CORE CORE CORE CORE CORE CORNIER K IS LESS THAN 1:0 CORE CORE CORE CORE CORE CORE CORNIER K IS LESS THAN 1:0 CORE COR		i			CORE	302		
40 IF (1PRINT.GE.2)50 TO 103 WRITE(6,20) NPKS 20 FORMAT(110,0) FERS OF PERKS OR VALLEYS IN THE IMPUT LOAD SPE CORE 10 FORMAT(6,22) WRITE(6,22)			1.NPKS		CORE	304	the land is specify and demandable property and the specify of the specific of the	
20 FORMAT(1H0,60HTHE NUMBER OF PEAKS OR VALLEYS IN THE IMPUT LOAD SPE CORE 10 TRUM = ,15//) WRITE(6,22) WRITE(6,22) WRITE(6,22) WRITE(5,22) WRI	04	IF (IPRINT 6	0.2.) 50 TO 103		CORE	305		
45 22 FORMAT(6,22) 45 22 FORMAT(60X,5HSIGMA/31X,4HSTEP,13X,714MAXIMUM,16X,7H4INIMUM,13X, CORE 9HCOUNTER K7) WRITE(6,25) (NSTEP (1),SIGMAX(1),SIGMIN(1),RNCYC(1), I = 1,NPKS) CORE 25 FORMAT(29X,15,10X,613,6,10X,F10,5) CONTINJE 50 5 SORT THROUGH THE LOAD SPECTRUM - PULL OUT THOSE PEAKS AND VALLEYS CORE C COUNTER K IS LESS THAM 1.0 C COUNTER K IS LESS THAM 1.0 C COURTER CORE		j ++	- NUMBER OF	YS IN THE IMPUT LOAD SP	i .	307		
1 9HGOUNTER K71 WRITE(5,25) (NSTEP (I),SIGMAX(I),RNGYG(I), I = 1,NPKS) GORE 25 FOFMAT(29X,I5,10X,E13.6,10X,E13.6,10X,F10.5) 103 CONTINJE 50 5 SORT THROUGH THE LOAD SPECTRUM - PULL OUT THOSE PEAKS AND VALLEYS GORE C COUNTER K IS LESS THAM 1.0 C COUNTER K IS LESS THAM 1.0 C COURTER COORE COORE 555 L = 0 CORE COR	54		! -	M, 16X, 7HHINIMUM, 13X,	CORE	310		
25 FORMAT(29X,15,10X,E13.6,10X,E13.6,10X,F10.5) 103 CONTINJE CONTINJE CORE COUNTER INCOMENTE LOAD SPECTRUM - PULL OUT THOSE PEAKS AND VALLEYS CORE COUNTER K IS LESS THAM 1.0 CORE CORE 55 L = 0 CORE			SIGMA.	CYC(I), I =	CORE CORE	311		
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	311 CONTINUE	CORE	376	
	NPKSN = NPKSN - JHAS	CORE	377	
115	S RANGE PAIR COUNT THE ADJUSTED LOAD SPECTRUM	CORE CORE	379	
	6000 T = 1 KB = 0	CORE	381	
	KMIN = 0	CORE	383	
120	MAX =	CORE	386	
	31 =	CORE	387	
125	(KB .NE. 0) GO TO 5	CORE	389	
	" 5	CORE	301	
	" "	CORE	303	
130	K9 = 1 G0 T0 1	CORE	395	
	5 X3 = SIGMAX(I) X4 = SIGMIN(I)	CORE	397	
135	203	CORE	399	
	0 0	CORE	401	
27	ON X	CORE	404 404	
140	= 1 0 = 1	CORE	405	•
	60 TO 415	CORE	408	
145	CALL	CORE	409	
	1. (10,10,30), KCYGEN	CORE	412	
	I = I + 1 IF (KMIN .NE. 1) 30 T	CORE	413	
150	TF (1 +LE NPKSN) GO TO 5 RESLER+1) = X1	CORE	415	
	" -	CORE	417	
155	INDEX (LR+2) = INDE	CORE	419 420	
	TO IF (AMIN .NE. 14 GO TO 35	CORE	421 422	
	12 I = I + 1 IF (I .LE. NPKSN) GO TO 31	CORE	424	
160	" "	CORE	425	
	1	CORE	428	
	INDEX (LE+2) = IND2	CORE	429	

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180	S = hx S	and the second second distribution of the second	CORE	577	de de la company de la communicación de la company de la c	
165	ħ		CORE	445	to any other management of the party of the	
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185 36 x = \$15 x v t t	. 0	Andread the second of the seco	CORE	6 4 4		
150	X3 = SX	And the state of t	CORF	450	and the second s	
1940	IND3 =		CORE	451	to be and the substitute of th	
190	XIXX	tradita makan menganyan dapambah banyumna sebugun sebugun bermangan bermanggi bermanggi bermanggi bangan bermanggi b	CORE	452		
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190	K 10 2	and a second to the composition of the company of t	CORE	+ c + d		
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265	K31 = 0	2.01 GO TO	CORE	.467		
205	TI CACACITY ENGLISHED ENGL	0.1	CORE	204		
205	G0 T2 402	:	CORF	470	gana in despitatore in other parameters and speciments are a strongers.	
210	205 401	•	CORE	471		
4.10 x3 = SI3 MAX(I) CORE X4 = SIGHIN(I) CORE IND3 = NSTEP(I) CORE CORE KMIN = 1 CORE KIND = 1 CORE KIND = 1 CORE CORE CORE CORE CORE CORE CORE CORE	705		CORE	274		
X4 = SIGNIN(I) IND3 = NSTEP(I) IND4 = TNO3 CORE KMIN = 1 CORE KJ1 = 0 CORE KIND = 1 CORE KIND = 1 CORE	•		CORE	474		
IND3 = NSTEP(I) IND4 = I NOS CORE KMIN = 1 CORE KJ1 = 0 CORE KIND = 1 CORE KIND = 1 CORE	(S = hX		CORE	475		
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KYAX = 0 KY1 = 0 CORE KIND = 1 ENCYC(I) = 1.0 CORE CORE CYCHO = AINT(RNCYC(I)+0.5) CALL DESIDE(X1,X2,X3,X4,KEY,I;CYCHO,KCYGEN) CORE CORE CORE CORE CORE CORE	1 1		CORE	4//		
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415 CYCNO = AINT(RNCYC(L)+0.5) CALL DESIDE(X1,X2+X3,X4,KEY+I,CYCNO,KCYGEN) GO TO 1000	K3 =		CORE	483		
CO TO 1000	CYC	-	CORE	484		
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	695 bbur FTN V3.U-3514 OPT=1	09/15/73	09.20.05. PAGE 5	1
2000	LMAX = .	CORE	486	
	IF (NCYNO .EQ. 0) 50 TO 5000	CORE	187	
225	RANGE PAIR COUNT OF RESIDUE SPECTRUMS	CORE	4 8 9 4 9 0	
O		CORE	491	
	CALL DEDRES(LRMAX,NGYNO) GO TO 2000	CORE	493	
. 230 5000	1	CORE	564	
3	COUNT THE LAST RESIDUE SPECTIUM - RANGE PAIR COUNTING WILL VIELD N	N CORE	264	
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375	GO TO 550 KMAX = KK	CORE	527 528	
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275	KG = KC + 1 NSTEP(K2) = KP	CORE	539	

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	DAMAGE FROM PLASTIC STRAINS= .22665690E-02
	DAMAGE PER THIS SET= .26509046E-02
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APPENDIX III

LIST OF COMPUTER PROGRAM SYMBOLS AND DEFINITIONS

A Coefficient of the x^2 term in the equation of a line on a constant life fatigue diagram where minimum stress is x and maximum stress is y. ($R = Ax^2 + Bx + C - y$)

AA An assigned value of +1. or -1.

AAA A stress used in the calculation of plastic strain.

ABDIF The absolute value of DIF.

ABM The absolute value of ASMAX or of ASMIN, as assigned.

ABMAX The absolute value of ASMAX.

ABMEAN The absolute value of ASMEAN.

ABMIN The absolute value of ASMIN.

ABR4 The absolute value of R(4).

ABR7 The absolute value of R(7).

ABS The name of a routine calling for the absolute value of a quantity.

AKT Stress concentration factor, K

ASE Array of values of ENN, for the plotting subroutine.

ASMAX The product (AKT) (STMAX)

ASMEAN The quantity (ASMAX + ASMIN)/2

ASMIN The product (AKT) (STMIN)

Array of values of ASMIN, for the plotting subroutine.

ASK Array of values of ASMAX, for the plotting subroutine.

AVSCIN Average value of SIGMIN over an interval.

AVSCMX Average value of SIGMAX over an interval.

B Coefficient of the x term. (See A.)

BBB A stress used in the calculation of plastic strain.

C The constant. (See A.)

COFMAN Inverse of the Coffin-Manson slope.

CYCINT The number of cycles in an interval.

CYCLES The calculated number of cycles expected to be indicated on a constant life fatigue diagram for the applied combination of maximum and minimum stress.

C1 The Residual Stress Relaxation Constant (See ENEP.)

DAM Damage.

DECK Decimal or real value of integer K after conversion.

DEL2 A portion of a least-squares-method solution.

DIF The difference between residual stress and equilibrium residual stress. (RES(I) - EQRES)

DO2 A portion of a least-squares-method solution.

DUMMY A variable used in the calculation of the number of cycles to be considered as an interval for relaxation determination.

ELMOD The elastic modulus.

EN The number of cycles from the beginning of the relaxation process to the end of the current interval.

ENEP The number of cycles required for overload residual stress

effect to return to within one-tenth of its original difference
from equilibrium conditions.

 $(N_{ep} = C1/(ABM)^{E1}(ABMEAN)^{E2})$

The number of applied cycles at a load level.

ENNCYC The ratio of the number of applied cycles to the number

of cycles to failure. (ENN/CYCLES)

EPSD ICF strain intercept.

EQRES Equilibrium residual stress.

EX An exponential function depicting the relaxation of residual stress.

EXP The name of a routine calling for the exponential value of a quantity.

EXPO An exponent. The power of 10 which indicates the number of cycles to failure.

E1 Residual stress Relaxation Exponents.

The name of a routine calling for integer-to-real conversion.

I A variable subscript.

IBLOCK The identifying number of a block the blocks being numbered consecutively from 1 to NBLOCK.

IFIX The name of a routine calling for real-to-integer conversion.

IN The number of steps input to the range pair counting subroutine.

IPRINT Value controlling the WRITE statements.

RAIN A counter.

Value controlling entry into the range pair counting subroutine.

ISTEP The identifying step number, the steps being numbered from 1 to NLEVEL.

ITYPE The identifying type number, the types being numbered from 1 to NTYPE.

J A variable subscript.

JA Value of +1 or 0, as assigned for branch determination.

JB Value of -1 or 0, as assigned for branch determination.

JJ An index variable.

JJJ An index variable.

JKL An index variable.

K An index variable.

KK An index variable.

KPMAX The number of steps output from the range pair counting subroutine.

L An index variable.

LMN An index variable.

M An index variable.

N An index variable with values of N=4-7 indicating the power of 10, and thus identifying a particular life cycle curve.

NBLOCK The total number of times to execute a block of loads.

NDECK The number of data decks to be run sequentially,

NFLAG - An integer used as a counter.

NFLAG2 An integer used as a counter.

MEYEL The total number of steps, or levels, of loads in a block.

NN A subscripted variable used to indicate which types of

loads are experienced in which blocks.

NTYPE The total number of different types.

PLSTRA Plastic strain.

R Residue term in damage calculation.

RES Residual stress.

RNCYC The number of cycles for a level after exitting the range

pair counting subroutine.

SIGMAX Maximum stress.

SIGMIN Minimum stress.

STMAX Maximum applied stress.

STMIN Minimum applied stress.

SUMDEL Summation of damage for a flight.

SUMENN Accumulated total of applied cycles. (Summation of ENN).

SUMNC Accumulated cycle ratio. (Summation of ENN/CYCLES).

SUMR Summation of R(N), N=4,7.

SUMRN Summation of nR(N), N=4,7. n=4.

SUMR2 Summation of $R(N)^2$, N=4,7.

SUMR2N Summation of nR(N), N=4,7. n=4.

SUMR3 Summation of R(N)³, N=4,7.

SUMR4 Summation of R(N)4, N=4,7.

TITLE1, TITLE2 Identification of the source of the SN data.

TLL Tensile load limit.

TM1, TM2 Material type.

TTYS One-fifth of tensile yield stress.

TYS Tensile yield stress.

T1,T2,T3,T4,T5,T6,T7,T8 Test identifying information.

X Variable equivalent to SIGMIN.

Y Variable equivalent to SIGMAX.

APPENDIX IV

FATIGUE LIFE INPUT DATA FLR SEVERAL MATERIALS

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	YIELD	STRAIN	INVERSE	LIFE,	S-N LI	FE COEFF	CIENTS	
MATERIAL	STRESS	INTERCEPT	OF SLOPE	10 ¹	A(I)	B(I)	C(I)	
			,					
2024-T4	58.	0.4	-1.836	4	0020	.2091	62.6	
				5	0032	.4366	51.4	
				6	0035	.6207	42.2	
			0.4	7	0042	.7003	36.1	
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2219-T851	55.	0.4	-1.836	4	0022	.2204	55.8	,
	•			5 .	0018	.3320	48.3	
•				. 6	0015	.4628	39.7	::
		• "		7	0024	.6420	31.7	
7075- T 6	72.	0.4	-1.836	4	0020	.2801	71.7	
		• ,		5	0022	.5154	56.3	***
	.•		٠.	6	0014	.6141	44.6	
•			• .	7	0013	.6838	38.1	
RQC-100	125.	0.54	-1.493	4	0.0	.2136	98.3	: •
				5	0.0	.2927	88.5	
		• •		6	0.0	. 3669	79.1	
	,			7	0.0	.4376	70.3	10
							40.0	
Man-Ten	55.	1.11	-1.667	4	0.0	.2257	63.5	
	: .	••		5	0.0	.3520	53.1	7. 1
				6	0.0	.4669	43.7	
				7	0.0	.5678	35.4	
2010 Ob 1	160	0.4	-1.836	4	0002	.2567	162.4	÷
4340 Stee1	160.	0.4	-T + 030	5	0007	.5248	126.9	
				6	0007	.5557	113.5	1
				7				
	**		(s. 1)	<i>'</i> .	0005	.5557	108.5	
T1-6-4	158.	0.4	-1.836	4	0009	.2368	154.2	
3. U T	****			5	0006	.4640	110.3	
				6	0000	.4650	88.9	
				7 .	.0001	.4752	84.2	
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